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ALERT SERVICE BULLETIN ASB No. AA-04107 Revision B

SUBJECT: Energy Attenuating Wire
P/Ns 157-13000-39, 157-13000-40, 157-43000-61

MODELS AFFECTED: Bell Helicopter Textron Model 206B with
Energy Attenuating Seat Installations
and Energy Attenuating Wires held as spares.

COMPLIANCE: This bulletin shall be complied within 150 flight
hours of receipt of this bulletin.

DESCRIPTION: This Alert Service Bulletin is being issued in
response to the determination that the energy
attenuating wire utilized in these seat kits may
not meet design specification resulting in the
possibility of degraded energy attenuation
performance during hard landings/crashes.

FAA/DER APPROVAL: The engineering design change represented by
this Alert Service Bulletin complies with
applicable Federal Aviation Regulations and is
FAA approved.

MANPOWER: Approximately 2.0 hours.

IF OWNERSHIP OF AIRCRAFT HAS CHANGED, PLEASE FORWARD THIS BULLETIN TO NEW OWNER

MATERIAL:

<u>Energy Attenuating Wire (provided)</u>	<u>Qty</u>
P/N 157-13000-39	2
P/N 157-13000-40	2
P/N 157-43000-61	2

No additional materials are required.

REQUIRED TOOLS:

No Special Tools required

WEIGHT AND BALANCE:

Not affected

PUBLICATIONS AFFECTED:

None affected

ACCOMPLISHMENT INSTRUCTIONS:

Determine either manufacture date or installation date of subject energy attenuating wires. Date of manufacture is shown on Parts Identification Label (Figure 1).

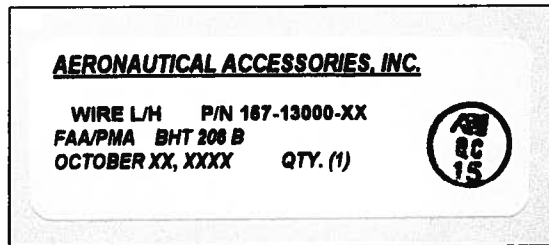


FIGURE 1 – Parts Identification Label

1. If wires are verified to have been manufactured outside the range of dates shown in table 1, or the date of the applicable Aeronautical Accessories invoice is prior to the dates, proceed to Section I, Step 13. If wire manufacture date is not verifiable or occurred within the range of dates shown, proceed to Section I, Step 1.
2. If wires are verified by written notation in the helicopter's log to have been installed prior to the dates shown in table 1 proceed to Section I, Step 13. If wire installation date is not verifiable or occurred within the range of dates shown, proceed to Section I, Step 1.

TABLE 1

EA Wire P/N	Dates (inclusive)
157-13000-39	Sept. 2001 – Sept. 2004
157-13000-40	Sept. 2001 – Sept. 2004
157-43000-61	Sept. 2002 – Sept. 2004

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SECTION I: ATTENUATING WIRE REPLACEMENT PROCEDURES
(Refer to Figures 2 thru 7)**NOTE**

Retain all removed hardware for reassembly and reinstallation.

NOTE

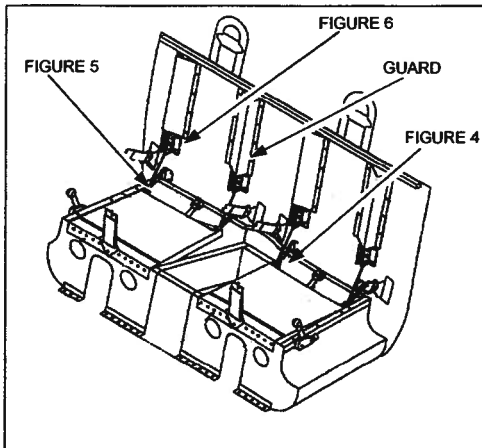
Seat backs, seat bases and trim panels are secured by hook and pile tape.

1. Remove seat and back cushions, which are secured by hook and pile tape, by carefully pulling at the corners of each cushion. Wrap seat and back cushions in protective material and set aside.
2. (Forward Crew Seats Only) Remove trim panels by grasping corners and carefully pulling away from seat assembly. Wrap trim panels in protective material and set aside.
3. Remove set screw (5) and jam nut (4). Lift seat high enough to clear bolt (1). Remove nut (8), washers (2), spacer (3), and bolt (1). Remove bracket (6).
4. Remove bolt (9, reference Figures 6 and 7), washers (10 and 11), and rollers (13) one at a time. Remove attenuating wire (7) from bracket (14 or 15).
5. Replace rollers (13) and bolts (9) if damage or corrosion is evident, or if rollers do not rotate freely on bolts.
6. Hold center roller (13) in position on new attenuating wire and slide upper end of attenuating wire into guard (Figure 2, 3). Position center roller against bracket (14 or 15) and insert bolt (9) with washers (10 and 11) in middle holes in bracket (14 or 15). Place top roller (13) in position and insert bolt (9) with washers (10 and 11). Place bottom roller (13) in position and insert bolt (9) with washers (10 and 11).
7. Install remaining washers (10) and nuts (12). Tighten nuts until rollers do not turn, then slightly loosen nuts so that rollers turn freely.
8. Install bracket (6) and align attenuating wire (7) with holes in seat and bracket (6). Install spacer (3), bolt (1), washers (2), and nut (8).
9. Install set screw (5) and jam nut (4). Tighten set screw (5) until set screw makes contact with lever (6). Tighten set screw $\frac{1}{4}$ turn more. Tighten jam nut (4) to 60 – 80 inch-pounds.
10. Install seat and back cushions by pressing into place.
11. (Forward Crew Seats Only) Install trim panels by pressing into place.
12. Replacement complete.
13. Make a notation in the helicopter's log indicating ASB No. AA-04107 has been accomplished.

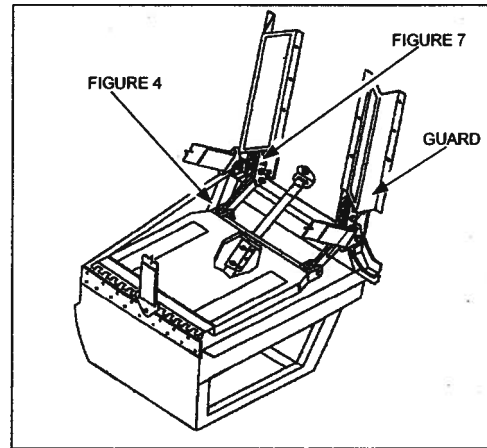
Any questions regarding this bulletin should be addressed to:

AERONAUTICAL ACCESSORIES, INC.
PRODUCT SUPPORT
1-800-251-7094

AERONAUTICAL ACCESSORIES, INC.



**FIGURE 2 – PILOT AND COPILOT
 ENERGY ATTENUATING
 SEAT INSTALLATIONS**



**FIGURE 3 –AFT OBSERVER
 ENERGY ATTENUATING
 SEAT INSTALLATIONS**

**Figures 4 thru 7
 Item Number Descriptions**

Item No.	Part Number	Description	Qty
1	NAS6604-16	Bolt	6
2	NAS1149D0416J	Washer	12
3	NAS43DD4-56FC	Spacer	6
4	MS35650-3252	Jam Nut	6
5	MS51966-129	Set Screw	6
6	157-13000-63 (L/H) 157-13000-64 (R/H)	Bracket	3 ea.
7	157-13000-39	Attenuating Wire – Copilot Seat	2
7	157-13000-40	Attenuating Wire – Pilot Seat	2
7	157-43000-61	Attenuating Wire – Aft Observer Seat	2
8	MS21042L4	Nut	6
9	NAS1304-16	Bolt	18
10	NAS1149D0463J	Washer	36
11	50Z12-9-2	Washer	18
12	MS21042L4	Nut	18
13	157-13000-37	Roller	18
14	157-13000-41/ -42	Bracket	2 ea.
15	157-13000-63/ -64	Bracket	1 ea.

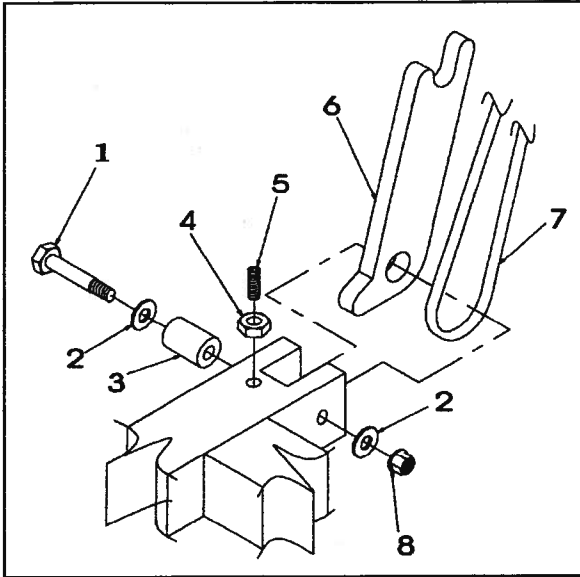


FIGURE 4 – COPILOT AND AFT OBSERVER ENERGY ATTENUATING WIRE SEAT ATTACHMENT

Note: Right side shown. Bolt (1) is always assembled pointing into seat; i.e., left side attachment hardware is opposite to what is shown in Figure 3 for Copilot and aft observer seats. For the Copilot seat, attenuating wire (7) is on the left of the bracket (6) on right and left sides. For the left side of the aft observer seat, attenuating wire (7) is opposite to what is shown in Figure 3.

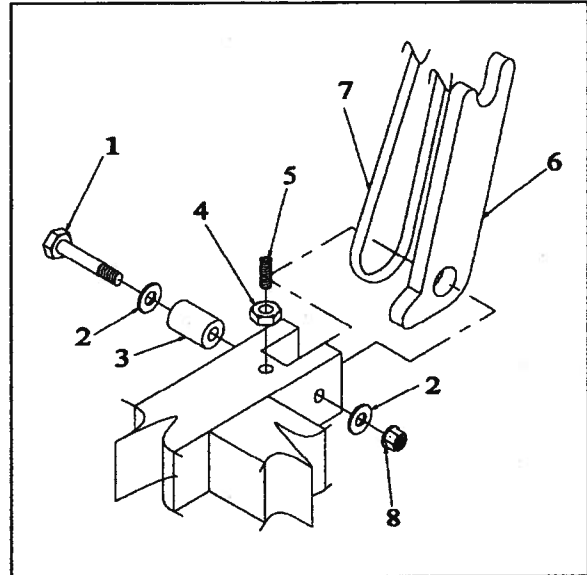


FIGURE 5 – PILOT ENERGY ATTENUATING WIRE SEAT ATTACHMENT

Note: Right side shown. Bolt (1) is always assembled pointing into seat; i.e., left side attachment hardware is opposite to what is shown in Figure 4. Attenuating wire (7) is on the right of the bracket (6) on both sides of the Pilot seat

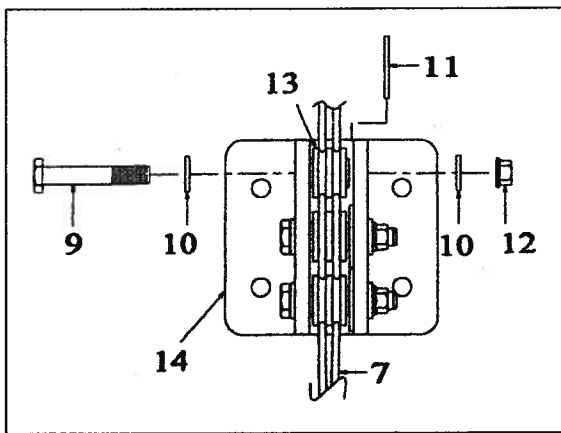


FIGURE 6 – PILOT AND COPILOT ENERGY ATTENUATING ROLLER BRACKET ASSEMBLY

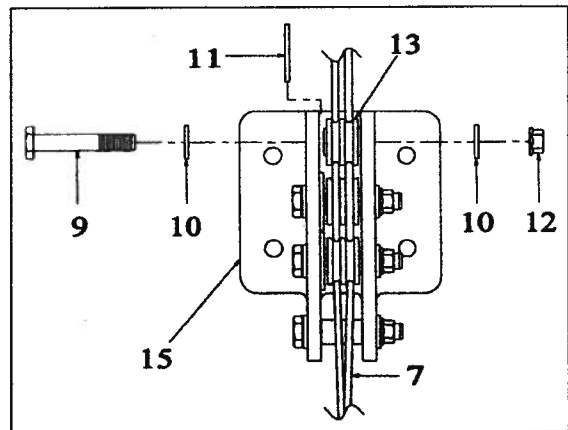


FIGURE 7 –AFT OBSERVER ENERGY ATTENUATING ROLLER BRACKET ASSEMBLY